**J-SSOD & [Satellite Name] Interface Verification Record**

**（For 10cm-sized Small Satellite）**

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| Satellite Developer Name | ; | [Defined by Satellite Developer] |
| Satellite Name | ; | [Defined by Satellite Developer] |
| P/N | ; | [Defined by Satellite Developer] |
| S/N | ; | [Defined by Satellite Developer] |

SIGNATURES / Satellite Development, Sponsor agency

 NAME DATE

 Satellite Development Team (Initiate)

 NAME DATE

 Satellite Development Team (Reviewed)

 NAME DATE

 Satellite Development Team (Approved)

 NAME DATE

 Sponsor Agency (Approved)

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **<< Mechanical Interface >>** |  |  |  |  |  |  |  |
| **1.** | **Satellite Type** | 　 | 1U / 2U / 3U /6U |  | 1U, 2U, 3U or 6U | N/A | N/A | Pare 2.1.2(1) |
|  |  |  |  |  |  |  |  |  |  |
| **2.** | **Width in -Z Plane** |  |  |  |  |  |  |  |
|  | a. | +X Plane | 　 | mm |  | 100.0+/-0.1mm | Measurement | Assembly Procedure(XX-AP-01) | Para 2.1.2(2)Figure2.1.2-1, 2a~2d |
|  | b. | +Y Plane | 　 | mm |  |
|  | c. | -X Plane | 　 | mm |  |
|  | d. | -Y Plane | 　 | mm |  |
|  |  |  |  |  |  |  |  |  |  |
| **3.** | **Width in +Z Plane** |  |  |  |  |  |  |  |
|  | a. | +X Plane | 　 | mm |  | 100.0+/-0.1mm | Measurement | Assembly Procedure(XX-AP-01) | Para 2.1.2(2)Figure2.1.2-1, 3a~3d |
|  | b. | +Y Plane | 　 | mm |  |
|  | c. | -X Plane | 　 | mm |  |
|  | d. | -Y Plane | 　 | mm |  |
|  |  |  |  |  |  |  |  |  |  |
| **4.** | **Rails Length** |  |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | mm(S/W or Spring) |  | [For Deployment S/W ] [For Separation Spring ] 113.5+/-0.1mm (1U) 111.5+/-0.1mm (1U) 227.0+/-0.1mm (2U) 225.0+/-0.1mm (2U) 340.5+/-0.3mm (3U) 340.5+/-0.3mm (3U)340.5+/-0.3mm (6U) 340.5+/-0.3mm (6U)or or366.0+/-0.3mm (6U) 366.0+/-0.3mm (6U) | Measurement | Assembly Procedure(XX-AP-01) | Para 2.1.2 (3)~(5)Para 2.1.3 (1)Figure2.1.2-1, 4a~4d |
|  | b. | Rail 2 | 　 | mm(S/W or Spring) |  |
|  | c. | Rail 3 | 　 | mm(S/W or Spring) |  |
|  | d. | Rail 4 | 　 | mm(S/W or Spring) |  |
|  |  |  |  |  |  |  |  |  |
|  |  |  |  |  |  |  |  |  |  |
| **5.** | **Rails Width** |  |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | x mm |  | Min 8.5 x 8.5 mm | Measurement | Assembly Procedure(XX-AP-01) | Para 2.1.3(3)Figure2.1.2-1, 5a~5d |
|  | b. | Rail 2 | 　 | x mm |  |
|  | c. | Rail 3 | 　 | x mm |  |
|  | d. | Rail 4 | 　 | x mm |  |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **6.** | **Rails Surface Roughness** |  |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | OK / NG |  | ≦ 1.6μm (Ra) (\*1) | Inspection (Machine work order, Inspection report, etc.) | Assembly Drawing(XX-AD-01) | Para 2.1.3(4)Figure2.1.2-1, 6a~6d |
|  | b. | Rail 2 | 　 | OK / NG |  |
|  | c. | Rail 3 | 　 | OK / NG |  |
|  | d. | Rail 4 | 　 | OK / NG |  |
|  |  |  |  |  |  | *(\*1) Arithmetic average of the roughness profile.* |  |  |  |
|  |  |  |  |  |  |  |  |  |  |
| **7.** | **Rails Edges Rounding** |  |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | OK / NG |  |  Min R1 mmor Min C1 mm | Inspection (Machine work order, Inspection report, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(5)Figure2.1.2-1, 7a~7d |
|  | b. | Rail 2 | 　 | OK / NG |  |
|  | c. | Rail 3 | 　 | OK / NG |  |
|  | d. | Rail 4 | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **8.** | **Rails Surface Area(+Z Plane)** |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | OK / NG |  | Min 6.5 x 6.5 mm | Inspection (Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(6) |
|  | b. | Rail 2 | 　 | OK / NG |  |
|  | c. | Rail 3 | 　 | OK / NG |  |
|  | d. | Rail 4 | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **9.** | **Rails Contact Length withJ-SSOD Rail Guides** |  |  |  |  |  |  |
|  | a. | Rail 1, +X | 　 | mm |  | ≧ 85.1mm (1U)≧ 170.3mm (2U)≧ 255.4mm (3U, 6U(+Z:340.5mm))≧ 274.5mm (6U(+Z:366.0mm)) | Analysis (Assessment based on Manufacture drawing, etc. is allowed.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(7) |
|  | b. | Rail 1, -Y | 　 | mm |  |
|  | c. | Rail 2, -Y | 　 | mm |  |
|  | d. | Rail 2, -X | 　 | mm |  |
|  | e. | Rail 3, -X | 　 | mm |  |
|  | f. | Rail 3, +Y | 　 | mm |  |
|  | g. | Rail 4, +Y | 　 | mm |  |
|  | h. | Rail 4, +X | 　 | mm |  |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **10.** | **Rail Surface Finish** |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | OK / NG |  | Anodized per MIL-A-8625 Type3 | Inspection (Machine work order, Inspection report, etc.) | Assembly Drawing(XX-AD-01) | Para 2.1.3(8) |
|  | b. | Rail 2 | 　 | OK / NG |  |
|  | c. | Rail 3 | 　 | OK / NG |  |
|  | d. | Rail 4 | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **11.** | **Clearance between Rail Edges & Main Structure (Z direction)** |  |  |  |  |  |  |
|  | a. | Rail 1, +Z | 　 | mm |  | ≧ 7mm | Inspection (Review of Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2), 2.1.4(1)(2)Figure 2.1.2-1, 11a~11d |
|  | b. | Rail 2, +Z | 　 | mm |  |
|  | c. | Rail 3, +Z | 　 | mm |  |
|  | d. | Rail 4, +Z | 　 | mm |  |
|  | e. | Rail 1, -Z | 　 | mm |  | ≧ 6.5mm | Inspection (Review of Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2), 2.1.4(1)(3)Figure 2.1.2-1, 11e~11h |
|  | f. | Rail 2, -Z | 　 | mm |  |
|  | g. | Rail 3, -Z | 　 | mm |  |
|  | h. | Rail 4, -Z | 　 | mm |  |
|  |  |  |  |  |  |  |  |  |  |
| **12.** | **Rails Perpendicularity against +Z Plane** |  |  |  |  |  |  |
|  | a. | Rail 1, +X | 　 | OK / NG |  | ≦ 0.2mm | Inspection (Machine work order, Inspection report, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2)Figure 2.1.2-1, 12a~12h |
|  | b. | Rail 1, -Y | 　 | OK / NG |  |
|  | c. | Rail 2, -Y | 　 | OK / NG |  |
|  | d. | Rail 2, -X | 　 | OK / NG |  |
|  | e. | Rail 3, -X | 　 | OK / NG |  |
|  | f. | Rail 3, +Y | 　 | OK / NG |  |
|  | g. | Rail 4, +Y | 　 | OK / NG |  |
|  | h. | Rail 4, +X | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **13.** | **Rails Perpendicularity against +Y Plane** |  |  |  |  |  |  |
|  | a. | Rail 1, +X | 　 | OK / NG |  | ≦ 0.2mm | Inspection (Machine work order, Inspection report, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2)Figure 2.1.2-1, 13a~13d |
|  | b. | Rail 2, -X | 　 | OK / NG |  |
|  | c. | Rail 3, -X | 　 | OK / NG |  |
|  | d. | Rail 4, +X | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **14.** | **Rails Parallelism to +Y Plane** |  |  |  |  |  |  |
|  | a. | Rail 1, -Y | 　 | OK / NG |  | ≦ 0.2mm | Inspection (Machine work order, Inspection report, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2)Figure 2.1.2-1, 14a~14b |
|  | b. | Rail 2, -Y | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **15.** | **Rail Edges Flatness on +Z Plane** |  |  |  |  |  |  |  |
|  | a. | Rail 1 | 　 | OK / NG |  | ≦ 0.2mm | Inspection (Machine work order, Inspection report, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.3(2)Figure 2.1.2-1, 15a~15d |
|  | b. | Rail 2 | 　 | OK / NG |  |
|  | c. | Rail 3 | 　 | OK / NG |  |
|  | d. | Rail 4 | 　 | OK / NG |  |
|  |  |  |  |  |  |  |  |  |  |
| **16.** | **Envelope (\*2)** |  *(\*2) Dynamic deformation shall be considered.* |  |  |  |
|  | a. | +X Plane | 　 | mm |  | ≦ 6.5mm | Measurement(or Inspection) | Assembly Procedure(XX-AP-01) | Para 2.1.4 (1)&(4)Figure 2.1.4-1, 16a~16d |
|  | b. | +Y Plane | 　 | mm |  |
|  | c. | -X Plane | 　 | mm |  |
|  | d. | -Y Plane | 　 | mm |  |
|  | e. | +Z Plane | 　 | mm |  | ≧ 0.5mm from rail surfaces (+Z). | Measurement(or Inspection) | Assembly Procedure(XX-AP-01) | Para 2.1.4 (1)&(2)Figure 2.1.4-1, 16e |
|  |
|  | f. | -Z Plane | 　 | OK / NG |  | No protrusion from rail surfaces (-Z). | Inspection | Assembly Procedure(XX-AP-01) | Para 2.1.4 (1)&(3)Figure 2.1.4-1, 16f |
|  |
|  | g. | Constraints on deployable components |  | OK / NG |  | Any deployable components shall be constrained by the satellite itself. The J-SSOD rails and walls shall not be used to constrain these deployables. | Review of Design | Assembly Procedure(XX-AP-01) | Para 2.1.4 (5) |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **17.** | **Mass Properties** | 　 |  |  |  |  |  |  |
|  | a. | Mass | 　 | Kg |  | 0.13～1.33kg/1U (1U,2U,3U)≦ 14kg (6U) | Measurement | Assembly Procedure(XX-AP-01) | Para 2.1.5(1) |
|  | b. | Ballistic Number | 　 | kg/m2 |  | ≦ 100 kg/m2 | Analysis | Structure Analysis Report(XX-SR-01) | Para 2.1.5(2) |
|  | c. | Center of Gravity | 　 | OK / NG |  | Within a sphere of 2 cm from the satellite geometric center. | Analysis(or Test) | Structure Analysis Report(XX-SR-01) | Para 2.1.5(3) |
|  | 　 |  |  |  |  |  |  |  |  |
| **18.** | **Separation Spring (1U & 2U Only)** |  |  |  |  |  |  |  |
|  | a. | Location | 　 | Option # |  | Option 1 or Option 2 | Inspection (Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.6(1)Figure 2.1.6-2, 18a |
|  | b. | Parts Number | 　 | OK / NG |  | IA P/N: 251D939002-1 | Inspection (Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.6(1) |
|  | c. | Positional Tolerance | 　 | mm |  | ≦ 0.3mm(Basis: 4.25mm from rail surfaces) | Inspection (Manufacture drawing, etc.) | Assembly Procedure(XX-AP-01) | Para 2.1.6(1)Figure 2.1.6-2, 18c |
|  |  |  |  |  |  |  |  |  |  |
| **19.** | **Accessibility** | 　 | OK / NG |  | Accessible thru Access Window at either -Y or +X plane if required after the installation into the J-SSOD Satellite Install Case. | Inspection (Manufacture drawing, etc.), Fit Check with J-SSOD | N/A | Para 2.1.7(1)&(2) |
|  |  |  |  |  |  |  |  |
| **20.** | **Structural Strength** |  |  |  |  |  |  |  |
|  | a. | Main Structure Strength |  | OK / NG |  | A satellite shall have a sufficient structural strength with a necessary safety margin through the ground operation, testing, ground handling, and on-orbit operations. | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.1.8(1) |
|  | b. | Rails Strength |  | OK / NG |  | Each rail shall have a sufficient structural strength with 46.6 N of a combined load of the preload and the spring load by the main spring. | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.1.8(2) |
|  |  |  |  |  |  |  |  |  |  |
| **21.** | **Stiffness** |  | Hz |  | Minimum fundamental frequency ≧ 100 [Hz] | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.1.9 |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **<< Electrical Interface >>** |  |  |  |  |  |  |  |
| **22.** | **Deployment Switches** |  |  |  |  |  |  |  |
|  | a. | Location |  | Option # |  | Option 1 or Option 2 | Inspection (Drawing order, etc.) | Assembly Procedure(XX-AP-01) | Para 2.2.1(1)Figure 2.1.6-2, 22a |
|  | b. | Function Test |  | OK / NG |  | Satellite shall not be activated when either of two switches remains depressed, i.e. 0.75mm max. from the rail standoff surface. | Function Test | Inhibit Function Test Report(XX-IFTR-01) | Para 2.2.1(2)Figure 2.2.1-1, 22b |
|  | c. | Stroke |  | mm |  | ≦ 2.0mm | Function Test | Inhibit Function Test Report(XX-IFTR-01) | Para 2.2.1(4)Figure 2.2.1-1, 22c |
|  | d. | Force |  | N |  | ≦ 3N | Function Test | Inhibit Function Test Report(XX-IFTR-01) | Para 2.2.1(5) |
|  |  |  |  |  |  |  |  |  |  |
| **23.** | **RBF Pin** |  |  |  |  |  |  |  |
|  | a. | Accessibility |  | -Ys / +Xs  |  | RBF pin shall be accessible thru Access Window at either -Ys or +Xs plane if required after the installation into the J-SSOD Satellite Install Case. | Inspection (Manufacture drawing, etc.), Fit Check with J-SSOD | N/A  | Para 2.2.2(1) |
|  | b. | Function Test |  | OK / NG |  | RBF pin shall cut all power to the satellite once it is inserted into the satellite. | Function Test | N/A  | Para 2.2.2(2) |
|  | c. | Envelope |  | mm |  | Protrudes ≦ 6.5mm | Inspection (Manufacture drawing, etc.), Fit Check | N/A  | Para 2.2.2(3) |
|  | d. | Tether |  | OK / NG |  | - Tether shall be attached to the RBF pin.- A satellite shall be loaded into the J-SSODSatellite Install Case with the tether attached. | Inspection (Manufacture drawing, etc.) | N/A  | Para 2.2.2(4) |
|  |  |  |  |  |  |  |  |  |  |
| **24.** | **Bonding** |  | -Ys / +Xs |  | The bonding interface shall be accessible thru Access Window at either -Ys or +Xs plane. | Inspection (Drawing order, etc.) | N/A  | Para 2.2.3(1) |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **<< Operational Requirements >>** |  |  |  |  |  |  |
| **25.** | **Maximum Stowage Duration** |  | OK / NG |  | Maximum stowage duration shall assume the max stowage duration may be about 1 year. | Review of Design (\*3) | The satellite is designed based on proper stowage duration | Para 2.3(1) |
|  |  |  |  |  |  |  |  |  |  |
| **26.** | **On-orbit Maintenance Limitation** |  | OK / NG |  | On-orbit maintenance limitation will not plan any activation, checkout, or maintenance after the delivery. | Review of Design (\*3) | No maintenance required | Para 2.3(2) |
|  |  |  |  |  |  |  |  |  |  |
| **27.** | **Cold Launch Requirements** |  | OK / NG |  | A satellite shall have a capability to survive in the cold launch environment (i.e. w/o power). | Review of Design (\*3) | No power required | Para 2.3(3) |
|  |  |  |  |  |  | *(\*3) It is allowed to describe a rationale in "Evidence document" instead of providing a document.*  |
| **28.** | **Minimum Time until Appendage Deployment & RF Radiation** |  |  |  |  |  |  |  |
|  | a. | Timer Setting  |  | OK / NG |  | ≧ 30 minutes | Function Test | Antenna Deployment and RF transmission Test Report(XX-AD&RFT-01) | Para 2.3(4)&(5) |
|  | b. | Function Test |  | OK / NG |  | Whenever either of two deployment switches is re-depressed, the timer shall be reset. | Function Test | Antenna Deployment and RF transmission Test Report(XX-AD&RFT-01) | Para 2.3(4)&(5) |
|  |  |  |  |  |  |  |  |  |  |
| **29.** | **Limitation of the satellite deployment window** |  | OK / NG |  | A satellite deployment window shall not be restricted by a satellite design. If limitation of the satellite deployment window exists, a satellite developer shall coordinate with JAXA. | Review of Design | No strict requirement | Para 2.3(6) |
|  |
| **<< Environmental Requirements >>** |  |  |  |  |  |  |
| **30.** | **Random Vibration and Acceleration** |  |  |  |  |  |  |  |
|  | a. | Quasi-static Acceleration |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.1(a)  | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.4.1 (a) |
|  | b. | Random Vibration |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.1(b)  | Test(Vibration Test Report) | Vibration Test Report(XX-VT-01) | Para 2.4.1 (b) |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **31.** | **On-orbit Acceleration** |  |  |  |  |  |  |  |
|  | a. | On-orbit Acceleration |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.2(a)  | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.4.2 (a) |
|  | b. | Acceleration induced by JEMRMS Emergency-Stop |  | OK / NG | A satellite shall assume the condition defined in the section 2.4.2(b)  | Analysis(Stress Analysis Report) | Structure Analysis Report(XX-SR-01) | Para 2.4.2 (b) |
|  |  |  |  |  |  |  |  |  |  |
| **32.** | **Pressure Environment** |  |  |  |  |  |  |  |
|  | a. | Pressure |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.3(a) | Review of Design (\*5) | The satellite is designed based on pressure condition | Para 2.4.3 (a) |
|  | b. | Depressurization Rate |  | m(\*4) | If V/A ≦ 50.8m (2000inch), analysis is not needed. If V/A > 50.8m (2000inch), Stress Analysis Report is needed. | Analysis(Stress Analysis Report, if necessary) | Interface Verification Record (XX-IVR-01) | Para 2.4.3 (b) |
|  |  |  |  | *(\*4) Please fill in V/A.* | *(\*5) It is allowed to write the purport of no problem in "Evidence document" instead of providing a document.*  |
|  |  |  |  |  |  |  |  |  |
| **33.** | **Thermal Environment** |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.4. | Review of Design (\*5) | The satellite is designed based on thermal condition | Para 2.4.4 |
|  |  |  |  |  |  |  |
| **34.** | **Humidity Environment** |  | OK / NG |  | A satellite shall assume the condition defined in the section 2.4.5. | Review of Design (\*5) | The satellite is designed based on humidity condition | Para 2.4.5 |
|  |
| **35.** | **Out-gassing** |  | OK / NG |  | Rating “A” materials shall be used for a satellite. | Inspection(MIUL, MUA) | MIUL(XX-MIUL-01) | Para 2.5 |
|  |  |  |  |  |  |  |  |  |
| **<< Safety Requirements >>** |  |  |  |  |  |  |
| **36.** | **Safety Assessment Analysis** |  |  |  |  |  |  |  |
|  | a. | On-orbit Safety |  | OK / NG |  | A satellite provider shall conduct safety analysis and submit SAR. Necessary inspections and tests for safety assessment shall be also conducted. | Analysis, Test, Inspection (Phase III approved SAR)  | Safety Assessment Report for Phase3(XX-SAR-02) | para 4.2.1 |
|  | b. | Launch Site & Vehicle Safety |  | OK / NG | A satellite provider shall submit ATV/HTV/KSC Form 100 check list for launch site & vehicle safety assessment. | Analysis, Test, Inspection (ATV/HTV/KSC Form 100 check list) | (If HTV used)Form 100 | para 4.2.1 |

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| **No.** | **Item** |  | **Results** |  | **Requirement** | **VerificationMethod** | **Evidence document(Document No)** | **Reference** |
| **37.** | **Envelope** |  |  |  |  |  |  |  |
|  |  | Contact surface of the deployable components |  | mm |  | If any deployable components make contact with the inside wall of the J-SSOD Satellite Install Case in their unintentional deployment, the contact surface of the deployable components shall have more than 1mm thickness. | Inspection | Assembly Procedure(XX-AP-01) | Para 2.1.4 (６) |
| **38.** | **RF** |  |  |  |  |  |  |  |
|  | (1) | Frequency and Current Limit |  | mA |  | If downlink frequency below 110 MHz is used, maximum current in the circuits shall not exceed 50 mA.  | Test | N/A | Para 2.2.4 (1) |
|  | (2) | Allowable RF Radiation Levels |  | uV/mHz |  | RF radiation levels shall not exceed values of Table 2.2.4-1. | Analysis | Safety Assessment Report for Phase3(XX-SAR-02) | Para 2.2.4 (2) |



R1/C1

1U/2U/3U : 100±0.1

6U : 226.3±0.1

(2ヶ所)

6U: Ha=340.5±0.3 or 366.0±0.3

6U: Ha=340.5±0.3 or 366.0±0.3

/C1



